

Three-Dimensional Digital Image Correlation of a Composite Overwrapped Pressure Vessel During Hydrostatic Pressure Tests

Duane M. Revilock, Jr.
Glenn Research Center, Cleveland, Ohio

John C. Thesken
Ohio Aerospace Institute, Brook Park, Ohio

Timothy E. Schmidt
Trilion Quality Systems, West Conshohocken, Pennsylvania

Bradley S. Forsythe
Honeywell Technology, Las Cruces, New Mexico

NASA STI Program . . . in Profile

Since its founding, NASA has been dedicated to the advancement of aeronautics and space science. The NASA Scientific and Technical Information (STI) program plays a key part in helping NASA maintain this important role.

The NASA STI Program operates under the auspices of the Agency Chief Information Officer. It collects, organizes, provides for archiving, and disseminates NASA's STI. The NASA STI program provides access to the NASA Aeronautics and Space Database and its public interface, the NASA Technical Reports Server, thus providing one of the largest collections of aeronautical and space science STI in the world. Results are published in both non-NASA channels and by NASA in the NASA STI Report Series, which includes the following report types:

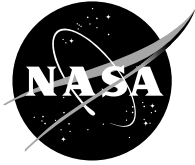
- **TECHNICAL PUBLICATION.** Reports of completed research or a major significant phase of research that present the results of NASA programs and include extensive data or theoretical analysis. Includes compilations of significant scientific and technical data and information deemed to be of continuing reference value. NASA counterpart of peer-reviewed formal professional papers but has less stringent limitations on manuscript length and extent of graphic presentations.
- **TECHNICAL MEMORANDUM.** Scientific and technical findings that are preliminary or of specialized interest, e.g., quick release reports, working papers, and bibliographies that contain minimal annotation. Does not contain extensive analysis.
- **CONTRACTOR REPORT.** Scientific and technical findings by NASA-sponsored contractors and grantees.

- **CONFERENCE PUBLICATION.** Collected papers from scientific and technical conferences, symposia, seminars, or other meetings sponsored or cosponsored by NASA.
- **SPECIAL PUBLICATION.** Scientific, technical, or historical information from NASA programs, projects, and missions, often concerned with subjects having substantial public interest.
- **TECHNICAL TRANSLATION.** English-language translations of foreign scientific and technical material pertinent to NASA's mission.

Specialized services also include creating custom thesauri, building customized databases, organizing and publishing research results.

For more information about the NASA STI program, see the following:

- Access the NASA STI program home page at <http://www.sti.nasa.gov>
- E-mail your question via the Internet to help@sti.nasa.gov
- Fax your question to the NASA STI Help Desk at 301-621-0134
- Telephone the NASA STI Help Desk at 301-621-0390
- Write to:
NASA Center for AeroSpace Information (CASI)
7115 Standard Drive
Hanover, MD 21076-1320



Three-Dimensional Digital Image Correlation of a Composite Overwrapped Pressure Vessel During Hydrostatic Pressure Tests

Duane M. Revilock, Jr.
Glenn Research Center, Cleveland, Ohio

John C. Thesken
Ohio Aerospace Institute, Brook Park, Ohio

Timothy E. Schmidt
Trilion Quality Systems, West Conshohocken, Pennsylvania

Bradley S. Forsythe
Honeywell Technology, Las Cruces, New Mexico

Prepared for the
2007 Semi Annual Conference and Exposition
sponsored by the Society for Experimental Mechanics
Springfield, Massachusetts, June 3–6, 2007

National Aeronautics and
Space Administration

Glenn Research Center
Cleveland, Ohio 44135

This report contains preliminary findings,
subject to revision as analysis proceeds.

Trade names and trademarks are used in this report for identification
only. Their usage does not constitute an official endorsement,
either expressed or implied, by the National Aeronautics and
Space Administration.

Level of Review: This material has been technically reviewed by technical management.

Available from

NASA Center for Aerospace Information
7115 Standard Drive
Hanover, MD 21076-1320

National Technical Information Service
5285 Port Royal Road
Springfield, VA 22161

Available electronically at <http://gltrs.grc.nasa.gov>

Three-Dimensional Digital Image Correlation of a Composite Overwrapped Pressure Vessel During Hydrostatic Pressure Tests

Duane M. Revilock, Jr.
National Aeronautics and Space Administration
Glenn Research Center
Cleveland, Ohio 44135

John C. Thesken
Ohio Aerospace Institute
Brook Park, Ohio 44142

Timothy E. Schmidt
Trilion Quality Systems
West Conshohocken, Pennsylvania 19428

Bradley S. Forsythe
Honeywell Technology
Las Cruces, New Mexico 88004

Abstract

Ambient temperature hydrostatic pressurization tests were conducted on a composite overwrapped pressure vessel (COPV) to understand the fiber stresses in COPV components. Two three-dimensional digital image correlation systems with high speed cameras were used in the evaluation to provide full field displacement and strain data for each pressurization test. A few of the key findings will be discussed including how the principal strains provided better insight into system behavior than traditional gauges, a high localized strain that was measured where gages were not present and the challenges of measuring curved surfaces with the use of a 1.25 in. thick layered polycarbonate panel that protected the cameras.

Introduction

The advent of high performance aramid and carbon fiber has enabled the evolution of filament wound pressure vessels capable of extreme energy storage capacity per unit mass. Starting in the 1960s and 70s, this potential was recognized by Johns and Kaufman (ref. 1), Lark (refs. 2 and 3) and Faddoul (ref. 4) at the NASA Lewis Research Center as a number of design and manufacturing studies began to investigate the technical feasibility of filament wound pressure vessels for space flight. Landes (ref. 5) and Ecord (ref. 6) published early work describing this technology with reported weight savings of 25 to 30 percent over comparable all metallic spherical vessels (ref. 7). Today (COPVs) are essential to numerous NASA power and environmental systems. The majority of older vessel overwraps are made of Kevlar (DuPont) 49/Epoxy Composites while the newer vessels have Carbon/Epoxy overwraps. The Kevlar 49 fiber overwrapped tanks are of particular concern due to their long usage and the poorly understood stress rupture process in Kevlar filaments. COPV's store high pressure gases for four different subsystems on the Space Shuttle Orbiter. There are 24 COPV's on each Orbiter ranging in diameters from 19 to 40 in. (fig. 1). These tanks were designed and developed in the late 1970s and most of them have been in service since delivery in the 1980s.

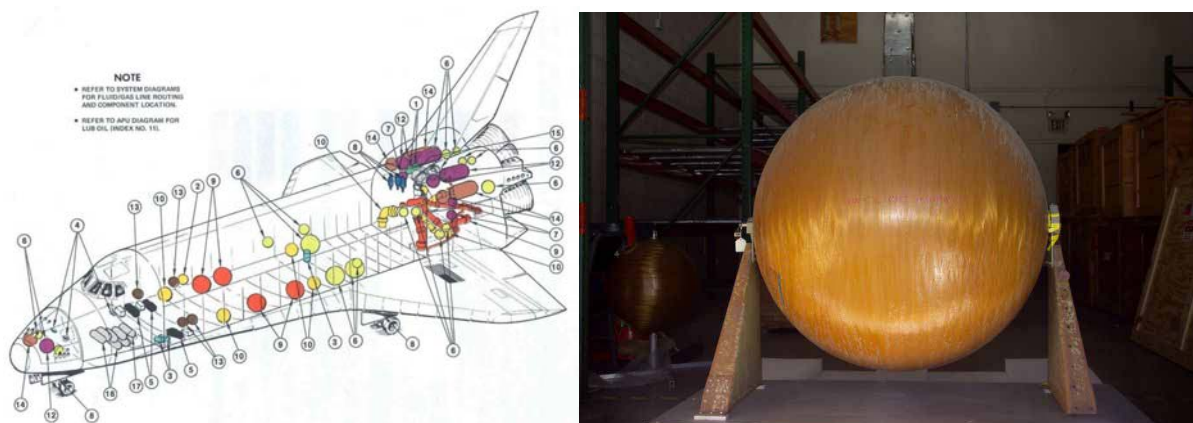


Figure 1.—Locations of composite overwrapped pressure vessels on a Space Shuttle Orbiter.

The vessels have a thin titanium liner with a Kevlar 49 composite overwrap. The overwrap is multiple layers and is designed to carry most of the load. Because stress rupture in Kevlar 49 gives no forewarning, Schmidt and Ecord (ref. 8) at the NASA Johnson Space Center (JSC) initiated an accelerated stress rupture test program to lead service hardware in actual time at pressure. The occurrence of burst events in that test program motivated the NASA Engineering and Safety Center to establish an Independent Technical Assessment of the (COPVs) used in NASA applications. That assessment showed that the Kevlar COPVs used on the Space Shuttle Orbiters were beyond their original certification of ten years and that the risk of a stress rupture, a catastrophic burst before leak failure mode, was greater than previously believed (ref. 9).

While existing long term data show that the stress rupture process in Kevlar fiber is a function of fiber stress, temperature and time, it is questionable whether the standard stress—rupture life representation of data may be used by itself for future life extension of NASA COPVs. A substantial contributor to the uncertainty is the presence of load sharing liners and complex manufacturing procedures such that the state of actual fiber stress in flight hardware and sub-scale test articles is not clearly known. As is the case with many ageing aerospace systems, the objective to extend flight certification for this hardware would benefit substantially from two concerted efforts:

1. Improve the understanding of the component's complex mechanical response, state of stress and deformation.
2. Improve the fidelity of the stress rupture lifing methods data base and use of the appropriate reliability framework for the stress rupture threat.

In an effort to obtain further insight into understanding the complex mechanical response in COPV's the NASA Johnson Space Center's White Sands Test Facility (WSTF) conducted ambient temperature hydrostatic pressurization testing of a 40 in. diameter COPV to understand the fiber stresses in the COPV components. This vessel was the same size as an Orbital Maneuvering System COPV, and was pressurized to its maximum operating pressure, design pressure, and to burst. The test vessel was instrumented with a combination of strain gauges, fiber optic Bragg grating sensors, acoustic emissions sensors, axial boss linear variable differential transformers, cable girth measurement instruments, eddy current sensors, and two high speed three-dimensional digital image correlation systems (DIC). The two digital image correlation systems used in the evaluation provided full field displacement and strain of the overwrap to understand the overall strain gradients in the vessel during each pressurization test. This paper will discuss some of the results obtained and the challenges of using the DIC on curved surfaces viewing through a polycarbonate shield.

Experimental Setup

The DIC system used in this test program was developed by GOM mbh of Braunschweig, Germany and utilizes a software package called ARAMIS. The ARAMIS software uses principles of three-dimensional image correlation photogrammetry that gives full-field displacement and strain measurements. The system requires spraying random high contrast dot patterns onto a sample; this pattern is then tracked in ARAMIS by thousands of unique correlation areas known as facets. The center of each facet is a measurement point that can be thought of as a three-dimensional extensometer. Arrays of them form in-plane strain rosettes. The facet centers are tracked in each successive pair of images, with accuracy up to one hundredth of a pixel. Figure 2 shows a 3/8 in. dot pattern used on the 40 in. vessel.

Two three-dimensional digital image correlation systems with Phantom v7.0 high speed cameras (Vision Research Corp. Wayne, NJ) were used in the tests, one focusing on the upper boss area and the other on the equator of the vessel (fig. 3). The high speed cameras are not standard with the DIC system but have been validated by the NASA Glenn Ballistic Impact lab during impact test studies for the Space Shuttle Program after the Columbia Accident (refs. 10 to 12). The digital high-speed cameras were used at a resolution of 512 by 512 pixels giving a recording speed range between 100 to 8,000 frames per second. The Phantom cameras act as a stereo pair to create a volume of area in which the ARAMIS software can take measurements. This volume varies with the angle of the cameras and lens choice. Camera setup consisted of 16-mm lenses with the cameras angled at 15.3° , giving a measuring volume of 1060 mm/875 mm/875 mm. To calibrate for this volume, the camera resolution was set at 800 by 600 pixels and a 1200-mm NIST-traceable calibration cross was rotated and moved in specified locations to calibrate the sensor. The cameras used IRIG time for synchronization, which provided accuracy within $1\ \mu\text{s}$ between images. Each DIC system was able to solve approximately $70\ \text{in.}^2$ of area on the vessel. Both DIC systems were used at WSTF during all of the pressurization cycles. The 40 in. COPV was filled with water and pressurized at 5 psi/sec and at 50 psi/sec to its proof pressure, at 50 psi/sec to its maximum operating pressure, and at 50 psi/sec to burst.



Figure 2.—COPV dot pattern painted on vessel for digital image correlation.



Figure 3.—Three-dimensional digital image correlation system setup.

Results

Due to the high pressures during testing, a 1.25 in. layered polycarbonate shield was bolted to the test chamber in front of the cameras to provide protection for the cameras and other test equipment. Since the DIC was not calibrated with the shield between the cameras and the vessel two tests were conducted at low pressures to measure its effect. The tests showed that the shield caused an increase in the maximum displacement readings. These higher readings were a result of the optical abnormalities in the shield and a magnifying effect. Figure 4 shows the X strain comparison at a point on the vessel with the same coordinate system with and without the shield indicating the higher strain readings.

Strain X Comparison

The vessel was instrumented with 35 strain gages to measure strains in the hoop direction of the fiber wrap at multiple locations around the vessel. For comparison purposes, one DIC system was aligned to view four of the mounted gages. Figure 5 shows a comparison of the hoop strain measured with the DIC system and the strain measured with the strain gages near the equator of the vessel during the burst cycle.

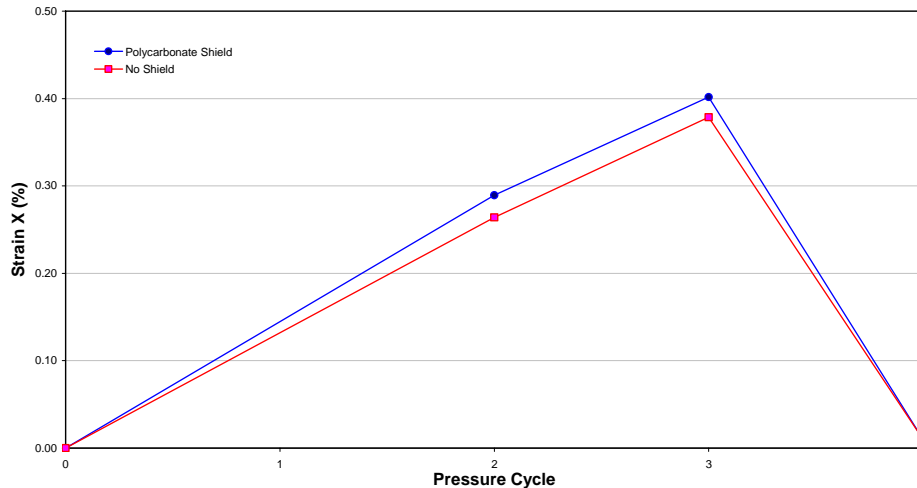


Figure 4.—X- strain comparison viewing through polycarbonate shield.

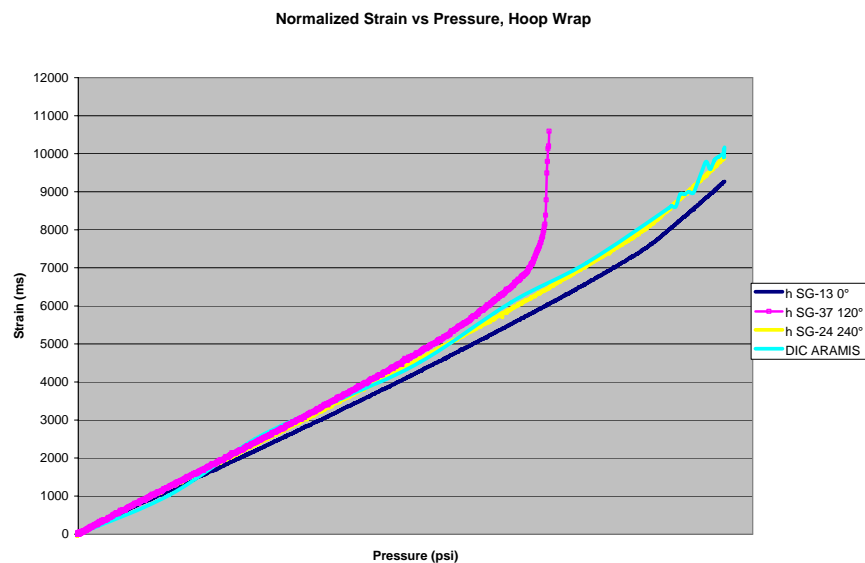


Figure 5.—Comparison of Digital Image Correlation Strain to Gages at the Equator of COPV.

Due to the curvature of the vessel, the effect of viewing through the polycarbonate shield, and visual interference from multiple instrumentation devices with cabling, the system was not able to compute consistent strains at the edges of the solved areas. Figure 6 shows the highly instrumented area the DIC is solving in and a snapshot of the noise floor from one of the tests. The full field strain x noise floor for most of the vessel is under 0.015 percent strain which is normal for this type of setup but it also indicates high and low localized strain regions at the edges of the solved area and at instrumentation locations. These high and low localized spikes occurred during each test and at various locations. To analyze this type of data a careful look at each point was needed to determine how accurate it was. During the pressurization cycles most of the spikes were located at points where the instrumentation cables were able to move. It was concluded that this movement was the source of the majority of the high noise floor regions. This made it difficult to determine the time and location of fiber failures during the burst cycle.

Strain Concentration

During the burst pressurization test at approximately 4.1 s before burst the DIC system showed a localized strain increase in the fourth wrap of the Kevlar in the upper portion of the boss area (figs. 7 and 8). These points (created with a new coordinate system with x being in the direction of the wrap) show strain progression from approximately 1.35 to 1.7 percent without visual fibers failing. Figure 9 shows the 26 percent increase in strain in this localized region which indicates a possible fracture in the titanium liner. This was supported by post test analysis which showed a crack in the liner at this location and an eddy current sensor located approximately 6 in. away indicating a thickening of the liner possibly due to stress relief as a result of the fracture. However, the closest strain gage approximately 2 in. away showed no indication of an increase in strain. This demonstrates the high degree of localization possible which may not be detected by functional strain gages unless one is located exactly at the fracture site.

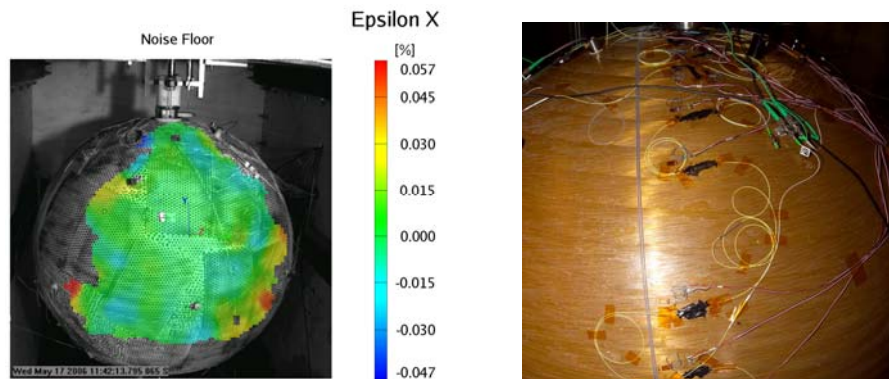


Figure 6.—Full field strain X showing hot spots at edges and instrumentation cable field in the same location.

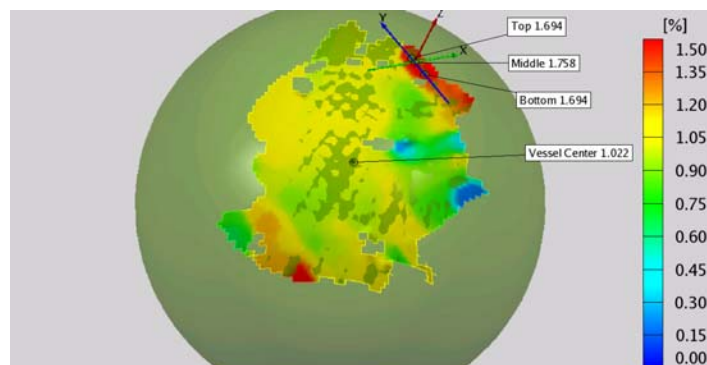


Figure 7.—Localized strain location at wrap four indicating a possible liner fracture. (Note full field strains will be slightly different due to coordinate system location).

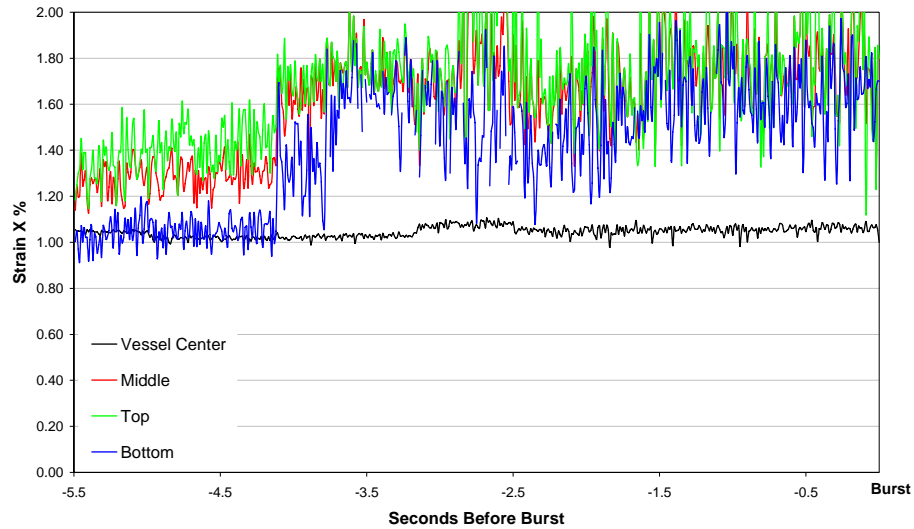


Figure 8.—Localized Point Strain Increase in Wrap Four Compared to Vessel Center. (Due to curved surface each point is created with a new coordinate system).

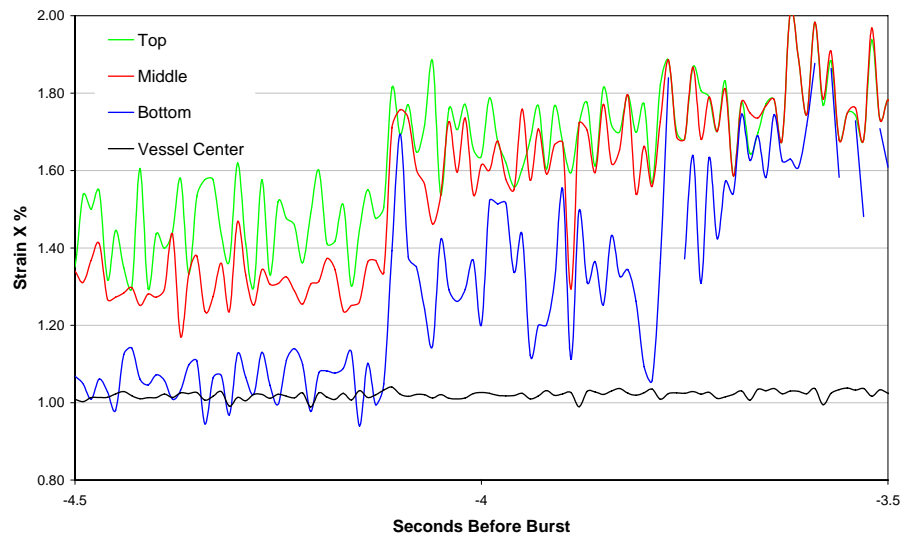


Figure 9.—Close up view indicating strain increase from top to bottom of wrap four.

Figures 8 and 9 also show more noise in the outer edges of the solved area than in the central region which is believed to be caused by the high viewing angle of the cameras to the curved surface and viewing through the polycarbonate shield. Even with this high noise the system is capable of showing trends of the strain increase. The system was also able to detect decreases in strain at the locations of fiber failure as the vessel approached burst. Figure 10 shows an example of a fiber failure during the burst test. The high speed video that supported the DIC system became very useful in visualizing the fiber breakage and matching it with the strain map to insure the data was not noise in the system.

Principal Strain Growth

Since the mounted gages could only provide strains in the hoop direction of the fiber wrap the DIC system became a valuable tool for measuring the principal strains. Figure 11 shows the principal strain noise floor for the DIC system indicating high noise up to 87 percent at the edges of the solved area and a strain offset of 30 to 40 percent closer in covering about 70 percent of the circumference.

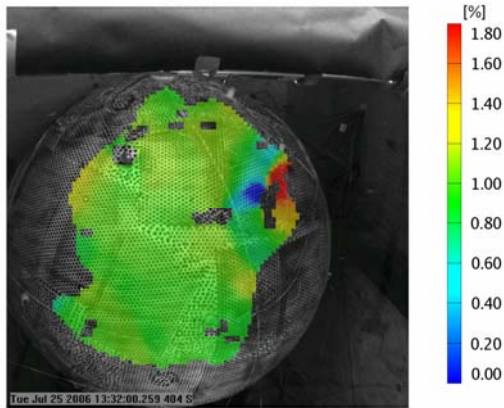


Figure 10.—Low strain reading in the blue area indicates fiber breakage.

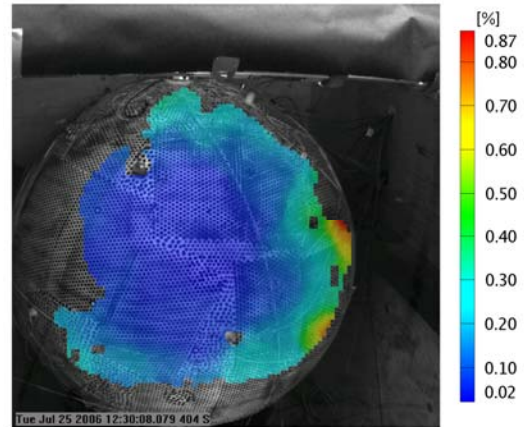


Figure 11.—Full field principal strain noise floor.

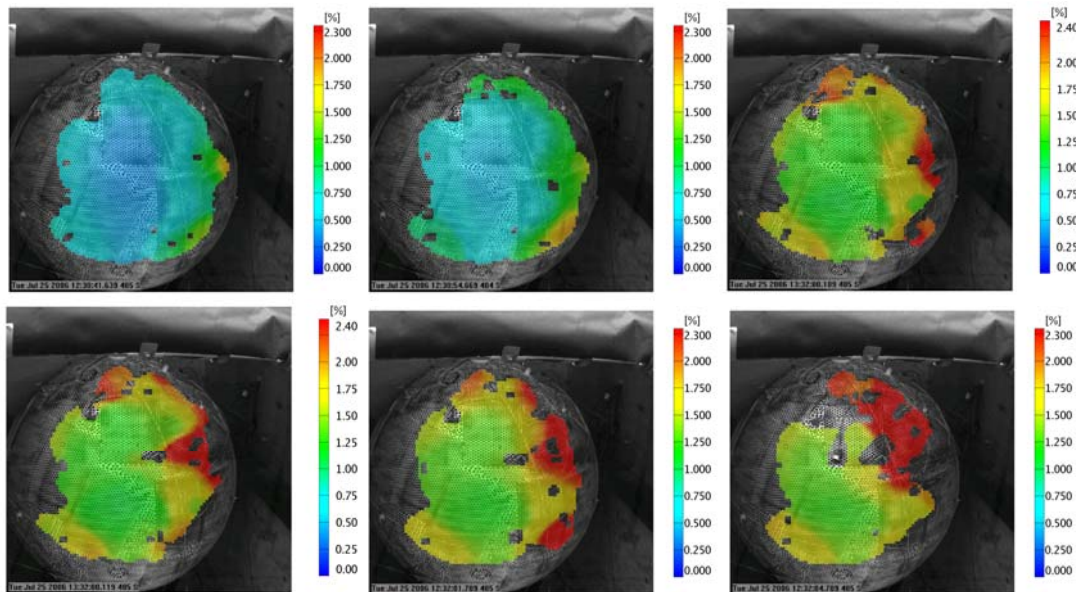


Figure 12.—Full field principal strain increase of the vessel during burst test.

This higher strain offset around the edges of the vessel is common for curved objects during similar tests and is generated by the high viewing angle of the cameras to the curved surface and by viewing through the polycarbonate shield. The data in the offset area consistently shows higher values but still is able to determine trends. Figure 12 shows a sequence of graphs of the principal strain during the burst test. The graphs indicate an increase in strain progressing from the right to the left with magnitudes above 2 percent which permeates past the offset area. This is consistent with post test analysis which showed that the vessel failed on the right side. Overall the DIC showed an increasing trend of principal strains on the right side of the vessel that mounted gages were unable to provide since they were only mounted in the one direction.

Conclusion

Two digital image correlation systems successfully provided accurate measurements of strain in a COPV during multiple pressurization tests which helped in the understanding of its complex mechanical

response during pressurization. Due to obtaining a large field of view and viewing through a polycarbonate shield a careful analysis was conducted to determine the validity of each area processed. Key results have shown a small increase in strain by viewing through a multi layered polycarbonate shield, an area showing a localized strain increase indicating a possible liner fracture, and a large growth in the principal strain during the burst test which mounted gages were not able to detect.

References

1. Johns, R.H. and A. Kaufman. 1966. Filament-Overwrapped Metallic Cylindrical Pressure Vessels, in AIAA/ASME Seventh Structures and Materials Conference, AIAA, 52–63.
2. Lark, R.F. 1973. “Filament-wound Composite Vessel Materials Technology,” NASA TMX–68196.
3. Lark, R.F., 1977. “Recent Advances in Lightweight, Filament-wound Composite Pressure Vessel Technology,” NASA TM 73699.
4. Faddoul, J.R. “Structural Considerations in Design of Lightweight Glass–Fiber Composite Pressure Vessels,” in Proceedings of the Second International Conference on Pressure Vessel Technology. Part I–Design and Analysis, ASME, 561–572, 1973.
5. Landes, R.E. 1973 “Glass Fiber Reinforced Metal Pressure Vessel Design Guide,” NASA CR-120917, Structural Composites Industries, Inc.
6. Ecord, G.M. “Filament wound pressure vessels with load sharing liners for Space Shuttle Orbiter applications” National Symposium and Exhibition on Bicentennial of Materials Progress, Los Angeles, CA, April 6–8, 1976, Jan. 1, 1976.
7. Landes, R.E. 1976. “Filament–Reinforced Metal Composite Pressure Vessel Evaluation and Performance Demonstration” Structural Composites Industries, Inc. Report SCI-75154, also NASA CR-134975.
8. Schmidt, W.W. and Ecord, G.M. “Static fatigue life of Kevlar aramid/epoxy pressure vessels at room and elevated temperatures,” AIAA PAPER 83-1328; June 1, 1983.
9. NESC Composite Overwrapped Pressure Vessel Independent Technical Assessment Final Report (in preparation).
10. J.M. Pereira, M. Melis, and D. Revilock Jr., “A Summary of the Space Shuttle Columbia Tragedy and the Use of Digital High Speed Photography in the Accident Investigation and NASA’s Return-to-Flight Effort,” SPIE 26th ICHSPP, Sept. 2004.
11. T. Schmidt, J. Tyson, D.M. Revilock Jr., J.M. Pereira, S. Padula II, M. Melis “Performance Verification of 3D Image Correlation Using High-Speed Cameras”. SEM 2005.
12. D. Revilock Jr., M. Melis, T. Schmidt, J. Tyson, K Galanulisb, “Full-field dynamic deformation and strain measurements using high-speed digital cameras”. SPIE 26th ICHSPP, Sept. 2004.

REPORT DOCUMENTATION PAGE				Form Approved OMB No. 0704-0188	
<p>The public reporting burden for this collection of information is estimated to average 1 hour per response, including the time for reviewing instructions, searching existing data sources, gathering and maintaining the data needed, and completing and reviewing the collection of information. Send comments regarding this burden estimate or any other aspect of this collection of information, including suggestions for reducing this burden, to Department of Defense, Washington Headquarters Services, Directorate for Information Operations and Reports (0704-0188), 1215 Jefferson Davis Highway, Suite 1204, Arlington, VA 22202-4302. Respondents should be aware that notwithstanding any other provision of law, no person shall be subject to any penalty for failing to comply with a collection of information if it does not display a currently valid OMB control number.</p> <p>PLEASE DO NOT RETURN YOUR FORM TO THE ABOVE ADDRESS.</p>					
1. REPORT DATE (DD-MM-YYYY) 01-08-2007		2. REPORT TYPE Technical Memorandum		3. DATES COVERED (From - To)	
4. TITLE AND SUBTITLE Three-Dimensional Digital Image Correlation of a Composite Overwrapped Pressure Vessel During Hydrostatic Pressure Tests				5a. CONTRACT NUMBER	
				5b. GRANT NUMBER	
				5c. PROGRAM ELEMENT NUMBER	
6. AUTHOR(S) Revilock, Duane, M, Jr.; Thesken, John, C.; Schmidt, Timothy, E.; Forsythe, Bradley, S.				5d. PROJECT NUMBER	
				5e. TASK NUMBER	
				5f. WORK UNIT NUMBER WBS 377816.06.03.02.10	
7. PERFORMING ORGANIZATION NAME(S) AND ADDRESS(ES) National Aeronautics and Space Administration John H. Glenn Research Center at Lewis Field Cleveland, Ohio 44135-3191				8. PERFORMING ORGANIZATION REPORT NUMBER E-16172	
9. SPONSORING/MONITORING AGENCY NAME(S) AND ADDRESS(ES) National Aeronautics and Space Administration Washington, DC 20546-0001				10. SPONSORING/MONITORS ACRONYM(S) NASA	
				11. SPONSORING/MONITORING REPORT NUMBER NASA/TM-2007-214938	
12. DISTRIBUTION/AVAILABILITY STATEMENT Unclassified-Unlimited Subject Categories: 18, 39, and 15 Available electronically at http://gltrs.grc.nasa.gov This publication is available from the NASA Center for AeroSpace Information, 301-621-0390					
13. SUPPLEMENTARY NOTES					
14. ABSTRACT Ambient temperature hydrostatic pressurization tests were conducted on a composite overwrapped pressure vessel (COPV) to understand the fiber stresses in COPV components. Two three-dimensional digital image correlation systems with high speed cameras were used in the evaluation to provide full field displacement and strain data for each pressurization test. A few of the key findings will be discussed including how the principal strains provided better insight into system behavior than traditional gauges, a high localized strain that was measured where gages were not present and the challenges of measuring curved surfaces with the use of a 1.25 in. thick layered polycarbonate panel that protected the cameras.					
15. SUBJECT TERMS Pressure vessels; Burst testing; Image correlation; Fiber composites; Composite wrapping					
16. SECURITY CLASSIFICATION OF:			17. LIMITATION OF ABSTRACT	18. NUMBER OF PAGES 14	19a. NAME OF RESPONSIBLE PERSON STI Help Desk (email: help@sti.nasa.gov)
a. REPORT U	b. ABSTRACT U	c. THIS PAGE U			19b. TELEPHONE NUMBER (include area code) 301-621-0390

